

a NASA facsimile reproduction

OF

N 62 - 14630

Reproduced From
Best Available Copy

AMPTIAC

B058132

REPRODUCED FROM MICROFORM

by the

Scientific and Technical Information Facility
DISTRIBUTION STATEMENT A
Approved for Public Release
Distribution Unlimited



20011116 016

N 62-14630

89.
FINAL REPORT
June 1962

MATERIAL EVALUATION FOR A
SUPERSONIC TRANSPORT PLANE

by

V. Weiss
R. Sell
C. Chave

for

Office of Research Grants and Contracts
CODE BG
National Aeronautics and Space Administration
Washington, D. C.

Contract No. NASr-43
Report Period: April 23, 1961-April 23, 1962

OTS PRICE

XEROX \$ 6.66 per page
MICROFILM \$ 2.00 per page

FINAL REPORT
June 1962

MATERIAL EVALUATION FOR A
SUPERSONIC TRANSPORT PLANE

by

V. Weiss
R. Sell
C. Chave

This report was produced under a sponsored contract. The conclusions and recommendations expressed are those of the Author(s) and are not necessarily endorsed by the Sponsor. Reproduction of this report, or any portion thereof, must bear reference to the original source and Sponsor.

SYRACUSE UNIVERSITY RESEARCH INSTITUTE
DEPARTMENT OF CHEMICAL ENGINEERING AND METALLURGY

Approved by:

Sponsored by:

ABSTRACT

4630 The screening program of materials being considered for a Supersonic Transport Plane conducted at Syracuse University during the period April 1961 through April 1962 is summarized in this report. The materials investigated are shown in Table I. Experimental results are shown in the form of graphs where the tensile strength, 0.2% yield strength, modulus of elasticity, elongation in percent, notch strength and the notch strength ratio are shown as a function of the test temperature. Evaluation of the alloys is also presented on the basis of notch strength ratio versus strength to density ratio. The titanium alloy Ti-6Al-4V annealed appears to be the most promising in this representation.

TABLE OF CONTENTS

INTRODUCTION	PAGE
EXPERIMENTAL RESULTS	1
AISI-301	4
PH 15-7 Mo	4
Ti-6Al-4V (RS-120A)	7
Ti-4Al-3Mo-1V (RS-115)	9
Ti-5Al-2.75Cr-1.25Fe (RS-140)	10
Ti-6Al-4V (RS-120A)	11
Ti-5Al-2.75Cr-1.25Fe (RS-140)	12
CONCLUSIONS	13
REFERENCES	14
TABLES	15
FIGURES	38

LIST OF TABLES

TABLES	PAGE
I MATERIALS	15
II CHEMICAL ANALYSIS	16
III NOTCH ROOT RADIUS CHANGES DURING EXPOSURE	17
IV, V TEST DATA AISI-301 COLD REDUCED 34%	18
VI, VII TEST DATA AISI-301 COLD REDUCED 51%	20
VIII, IX TEST DATA AISI-301 COLD REDUCED 60%	22
X, XI TEST DATA PH 15-7 Mo RH 1050	24
XII, XIII TEST DATA PH 15-7 Mo CH 900	26
XIV, XV TEST DATA Ti-6Al-4V (RS-120A) ANNEALED	28
XVI, XVII TEST DATA Ti-6Al-4V (RS-120A) SOLUTION TREATED AND AGED	30
XVIII, XIX TEST DATA Ti-4Al-3Mo-1V (RS-115) ANNEALED	32
XX, XXI TEST DATA Ti-5Al-2.75Cr-1.25Fe (RS-140) ANNEALED	34
XXII, XXIII TEST DATA Ti-5Al-2.75Cr-1.25Fe (RS-140) SOLUTION TREATED AND AGED	36

LIST OF FIGURES

FIGURES		PAGE
1	SPECIMEN DESIGN	38
2, 3	GRAPHIC DATA AISI-301 COLD REDUCED 34%	39
4, 5	GRAPHIC DATA AISI-301 COLD REDUCED 51%	41
6, 7	GRAPHIC DATA AISI-301 COLD REDUCED 60%	43
8, 9	GRAPHIC DATA PH 15-7 Mo RH 1050	45
10, 11	GRAPHIC DATA PH 15-7 Mo CH 900	47
12, 13	GRAPHIC DATA TI-6AL-4V (RS-120A) ANNEALED	49
14, 15	GRAPHIC DATA TI-6AL-4V (RS-120A) SOLUTION TREATED AND AGED	51
16, 17	GRAPHIC DATA TI-4AL-3Mo-1V (RS-115) ANNEALED	53
18, 19	GRAPHIC DATA TI-5AL-2.75Cr-1.25Fe (RS-140) ANNEALED	55
20, 21	GRAPHIC DATA TI-5AL-2.75Cr-1.25Fe (RS-140) SOLUTION TREATED AND AGED	57
22, 23	GRAPHIC DATA NOTCH STRENGTH RATIO VS STRENGTH/DENSITY	59

INTRODUCTION

This final report represents a summary of all data obtained to date at Syracuse University in the Screening Program of Materials being considered for a MACH III Transport Plane. During the course of this investigation three (3) quarterly progress reports have been submitted. (1, 2, 3)

To date, data on five materials listed below have been investigated:

- AISI-301 Stainless, 34%, 51% and 60% - Cold Reduction
- PH 15-7 Mo, RH 1050 and CH 900
- TI-6AL-4V (RS 120A) Annealed, Solution Treated and Aged
- TI-4AL-3Mo-1V (RS-115) Annealed
- TI-5AL-2.75Cr-1.25Fe (RS-140) Annealed, Solution Treated and Aged

The heat treating conditions and the chemical composition of these materials are listed in Tables I and II respectively.

The test program was primarily designed to provide a comparative evaluation of materials on the basis of tensile and notch tensile properties prior and subsequent to a 1,000 hour exposure under stress at 650°F. The exposure stress was chosen on the basis of preliminary design information on the expected service loads of the airplane. Steels and super-alloys were exposed under a stress of 40 ksi, titanium alloys under a stress of 25 ksi. The test specimens are illustrated in Fig. 1. All sheet materials had a thickness of 0.025 in. The tensile specimen had a width of .375 in. and a notch tensile specimen had a gross section width of 1 in. and a net section width of 0.705 in., the notch root radius being less than .001 in.

In addition, edge cracked specimens were tested for selected conditions in order to evaluate the influence of notch sharpness. Tensile tests and notch tensile tests were conducted at -110F, 75F, 350F, 650F and 800F, prior to exposure and at -110F, 75F and 650F subsequent to the above described 1,000 hour exposure under stress.

The tensile and notch tensile tests were conducted on a 50,000 lbs. capacity Baldwin Universal Testing Machine equipped with a micro-former stress-strain recorder. For the tests at -110F specimens were immersed in a solution of dry ice and acetone, room temperature (75F) tests were conducted in air, for elevated temperature tests the specimen was tested inside a Marshall furnace. Stress-strain curves were obtained for all tests on smooth specimens to permit the determination of the modulus of elasticity and the .2% yield strength as a function of test temperature and exposure treatment.

The exposure treatment was conducted in stress rupture machines converted for this purpose to accommodate two or three specimens within the constant temperature zone of the furnace.

In the preparation of notch tensile specimens special attention was given to an exact measurement of the root radii both prior and subsequent to exposure. All specimens having notch root radii in excess of .001 in. were rejected. A Sonntag Universal Fatigue Testing Machine was converted to allow the manufacture of edge cracked specimens from a 10% starter notch. With the exception of Ti-5Al-2.75Cr-1.25Fe (RS 140) in the solution treated and aged condition all materials were suitable to fatigue cracking by this method.

The following discussion of the experimental results includes all data obtained to date. Additional tests under the same program are now in progress on some materials contained in this report. The results will be discussed at a later date. In general the future effort consists of a continuation and amplification of the screening program. In the latter phase transition temperature studies are under way on Ti-6Al-4V, (Annealed, Solution Treated and Aged), PH 11.5-7 Mo, (RH 1050, CH 900) and Ti-8Al-1Mo-1V in the duplex annealed condition. The effect of exposure under stress on the ductile-to-brittle transition temperature will also be investigated on the more promising alloys. In addition a follow-on program is being initiated on wider specimens for the more promising alloys. A study on the effect of stress concentration factor in which specimens having various stress concentration factors between 1.5 and that corresponding to a natural crack has also been initiated.

P 3

P 2

EXPERIMENTAL RESULTS

In the following discussion of the experimental results each alloy will be discussed separately. An overall comparison of the relative rating of all alloys will be given in the Conclusions. The data are presented as a function of test temperature and fall generally in two categories: properties obtained from smooth specimens such as tensile strength, 0.2% offset yield strength, elongation in 2 in. and modulus of elasticity, and the properties obtained from notched specimens namely notch strength (on notched and cracked specimens) and notch strength ratio (notch strength/tensile strength, both taken at the test temperature indicated.) Longitudinal properties are represented by open symbols, transverse properties by solid symbols, properties prior to exposure by circles, properties subsequent to exposure by triangles. In some cases crack data were obtained and are represented by squares. Whenever data were obtained from an exposure treatment in which the specimen was subjected to the exposure temperature of 650° only and not simultaneously subjected to a load (no-load exposure) the data are represented by inverted triangles.

AISI-301 (Tables IV through IX)

Three strength levels of this alloy were investigated as obtained by a 34%, a 51% and a 60% cold reduction, namely 175 ksi, 220 ksi and 225 ksi respectively. The results are presented in Figs. 2 to 7.

Fig. 2 shows the smooth properties of AISI-301 cold reduced 34%. The tensile and yield strengths show a slight increase at all heat temperatures

of directionality causing lower transverse properties at all test temperatures, but particularly at -110F and at 75F. The elongation drops rapidly with increasing test temperature from approximately 18% at -110F to less than 4% at and above 350F. Adverse transverse properties are again observed below 350F. The exposure treatment causes a considerable loss in elongation at -110 and 75F in the transverse direction and at 75F in the longitudinal direction. Specimens subjected to a no-load exposure gave the same results as those exposed under stress.

The notch properties of AISI-301, cold reduced 34% are illustrated in Fig. 3. The notch strength continuously decreases with increasing test temperature. The transverse notch strength is generally 10% below the longitudinal notch strength. At -110F and at 75F exposure causes an increase in notch strength. No effect of exposure is observed at 650F and 800F (No-load exposure yielded approximately the same results as exposure under stress. The results obtained from fatigue cracked specimens are identical to those obtained from sharply notched specimens. The notch strength ratio increases from -110F to 75F and subsequently decreases with increasing test temperature. Exposure generally causes a loss in notch strength ratio except at -110F where the reverse effect is observed. A minimum notch strength ratio of 0.6 was observed at 650F subsequent to exposure in the transverse direction.

Fig. 4 shows the smooth properties of AISI-301 cold reduced 51%.

Tensile, yield strength and elongation decrease with increasing test

temperature in tensile and

yield strength at all test temperatures both in the longitudinal and in the transverse direction. This increase is approximately 20%. The elongation again drops rapidly from 18% at -110F in the longitudinal direction to less than 4% at 75F in both the longitudinal and the transverse direction. The exposure treatment causes a more than 50% loss in the elongation at -110F. The tensile strength in the transverse direction is slightly higher at -110F but no effect of directionality is observed on the tensile strength for other test temperatures. The yield strength in the transverse direction is below that in a longitudinal direction at -110F and 75F. The elongation and the transverse direction at -110F is also considerably below that for the longitudinal direction.

The notch properties are illustrated in Fig. 5. Transverse properties are always considerably below longitudinal properties, particularly at -110F and 75F where the notch strength in the transverse direction is almost 40% below the notch strength in the longitudinal direction. The notch strength ratio versus test temperature plot reflects the effect of the exposure treatment on the smooth properties. In all instances the notch strength ratio is decreased by the exposure treatment. The minimum notch strength ratio of .35 is observed at 650F subsequent to exposure in the transverse direction. →

Fig. 6 shows the smooth properties of AISI-301 cold reduced 50%. The experimental results are similar to those obtained on AISI-301 cold reduced 51% in that both tensile and yield strength are increased by the exposure treatment, particularly at -110F and room temperature. The elongation in

2% at room temperature. The elongation in the transverse direction and in both test directions subsequent to exposure never exceeds 4% for any test temperature investigated.

The notch properties, Fig. 8, decrease with increasing test temperature. Exposure has little effect but a marked difference exists between longitudinal and transverse properties, the latter being approximately 25% lower. The notch strength ratio also decreases with increasing test temperature and reflects the tensile properties as well as the directionality. Minimum notch strength ratios of 0.5 are observed at -110 and +650F subsequent to exposure when tested in the transverse direction. All test results obtained on fatigue cracked specimens of 301 stainless 34, 51 and 60% cold reduced agree with those obtained from machined notches.

PH 15-7 Mo (Tables X through XIII)

The results on PH 15-7 Mo in the condition RH 1050 are illustrated in Figs. 8 and 9. Both tensile and yield strength continuously decrease with increasing test temperature. Exposure as well as direction of testing seem to have little effect on the tensile and yield properties. The elongation decreases from approximately 5% at -110 to a minimum of less than 3% at 350F then increases with test temperature to approximately 4%.

Elongation shows both the effect of directionality as well as of exposure. In the region between 75F and 650F the elongation in the transverse direction is below that in the longitudinal direction. Exposure under stress seems to cause a slight increase in elongation of -110F and at 650F in the longitudinal direction while a loss is observed in the trans-

97

96

8

exposure indicate a loss in elongation at -110F in both directions and in the transverse direction at 75F.

The notch properties of PH 15-7 Mo, RH 1050, are illustrated in Fig. 9.

Both the effects of exposure and directionality are observed, the notch strength in the transverse direction being always considerably below that in the longitudinal direction. A sharp drop in notch strength due to exposure is observed at 75F. Notch strength data from fatigue cracked specimens lie slightly above those for machined notches at -110 and 75F.

No load exposure points were obtained at -100F and 75F. There appears to be little difference between the notch strength obtained prior to exposure and subsequent to a no load exposure. The notch strength ratio versus test temperature curves reflect the complexity of the notch strength versus test temperature curves and show a minimum at -110F of approximately 0.4. This low notch strength ratio was observed at -110F in the transverse direction prior to exposure and subsequent to exposure, the longitudinal direction subsequent to exposure and the transverse direction subsequent to a no load exposure.

The smooth properties of PH 15-7 Mo, in the condition CH 900 are represented in Fig. 10. To date only data prior to exposure are available.

Both tensile strength and yield strength decrease with increasing testing temperature. Little effect of directionality is observed except a slight increase in the transverse elongation at -110F. Over the entire range of testing temperature the elongation did not exceed 4%. The notch data on PH 15-7 Mo in the condition of CH 900 are represented in Fig. 11. Again

sharply with increasing testing temperature. Considerably lower notch strength values are observed in the transverse direction. This is reflected in the notch strength ratio values obtained namely .7 to .8 for the longitudinal direction and .35 to .45 for the transverse direction over the entire range of test temperatures investigated.

Specimens of PH 15-7 Mo both in the condition RH 1050 and CH 900 are now being exposed and the results on the effect of exposure on both notched and smooth qualities will be reported in the near future.

Ti-6Al-4V (RS-120A) (Tables XIV through XVII)

The experimental results of Ti-6Al-4V (RS-120A) in the annealed condition are illustrated in Figs. 12 and 13. Little effect of exposure treatment and directionality is noted on the tensile strength and the yield strength. The elongation within the range of test temperatures is between 6 and 13% showing a minimum of approximately 6% in the longitudinal direction prior to exposure at -110 and in the transverse direction prior to exposure at 800F. The notch strength, Fig. 13, decreases nearly linearly with increasing test temperature and again shows little effect of directionality or exposure. The notch strength ratio increases from approximately 0.8 at -110F to 1.0 at room temperature. The apparent susceptibility of this material to fatigue cracks should be emphasized. A minimum notch strength ratio .6 was observed on a fatigue crack specimen tested at -110F in the transverse direction.

Figs. 14 and 15 show the smooth and notch properties of Ti-6Al-4V (RS-120A) in the solution treated and aged condition. Tensile and yield strength decreases with increasing test temperature showing no effect of exposure or directionality. Exposure causes a significant loss in

elongation of specimens tested in the transverse direction, especially at -110F and 75F, where the elongation is reduced from approximately 7% prior to exposure to approximately 2% subsequent to exposure. No-load exposure data fall between the data obtained prior to exposure and subsequent to exposure under stress.

The notch properties, Fig. 15, show an increase in notch strength and notch strength ratio with testing temperature between -110F and 350F with a subsequent decrease with further increase in temperature. The effect of exposure and test direction is negligible, especially with respect to the notch strength ratio, as compared to the effect of test temperature. The minimum notch strength ratio .4 was observed on a fatigue crack specimen tested at -110F in the transverse direction.

Ti-4Al-3Mo-IV (RS 115) (Tables XVIII and XIX)

Figs. 16 and 17 show the smooth and notched properties obtained on Ti-4Al-3Mo-IV (RS-115) in the annealed condition. To date tests have only been performed on the material prior to exposure. Both tensile and yield strength gradually decrease with increasing test temperature. No effect of directionality is observed. The elongation increases from 4% to -110 to approximately 3% at 200F and subsequently decreases again to approximately 3% at 800F. The elongation measured in the transverse direction is lower than that measured in the longitudinal direction at 350F and 800F.

The notch properties, Fig. 17, increase between -110 and 200F and subsequently decrease with increasing testing temperature. Little effect of directionality is observed. The notch strength ratio increases through-

beyond unity at and above 200F. Specimens of this alloy are now being exposed and the results will be reported in the near future.

Ti-5Al-2.75Cr-1.25Fe (RS-140) (Tables XX through XXIII)

Figs. 18 and 19 show the smooth and notch properties of Ti-5Al-2.75Cr-1.25Fe (RS-140) in the annealed condition. The tensile strength and the .2% yield strength, Fig. 18, continuously decrease with increasing test temperature and show little effect of directionality and exposure. The elongation varies between 6 and 14% having a minimum at -110F and 600F in the longitudinal direction.

The notch strength, Fig. 19, increases between -110F and 75F where it reaches a maximum and then decreases nearly linearly as the test temperatures increase to 800F. The effect of directionality only observed at 75F, where the specimens tested in the transverse direction show slightly higher notch strength values. The exposure treatment seems to have little effect on the notch strength. The notch strength ratio increases from approximately .7 to 1.0 between -110F and 75F and remains nearly constant or above 1.0 as the test temperature is further increased to 800F. Little effect of exposure is noted.

Figs. 20 and 21 show the tensile and notch tensile properties obtained on RS-140 in the solution treated and aged condition. A general decrease in tensile and yield strength coupled with a slight increase in elongation is observed with increasing testing temperature from -110F to 800F, Fig. 20. No effect of exposure or directionality is observed on the tensile strength or yield strength. Minimum elongation values of approximately 2% are observed both in the longitudinal and transverse direction prior to sub-

10

11

12

sequent to exposure at -110°F. The gradual increase in elongation with increasing test temperature is less for specimens subjected to the exposure treatment.

The notch properties between -110 and 800°F are shown in Fig. 21.

The notch strength increases continuously with increasing testing temperature. Exposure causes a considerable reduction in notch properties at -110 and 75°F. Minimum notch strength ratio of .2 observed on specimens tested both in the longitudinal and transverse direction subsequent to exposure at -110°F. Only at testing temperatures above 350°F does a notch strength ratio exceed .6 for all conditions tested.

NOTCH ROOT RADIUS CHANGES DURING EXPOSURE

The effect of the exposure treatment on the notch geometry was determined on selected specimens of Ti-6Al-4V, solution treated and aged and PH 15-7 Mo, RH 1050. The results are shown in Table III. A slight increase in notch root radius was observed in one Ti-6Al-4V (transverse) specimen and both PH 15-7 Mo specimens. This increase is between 0.00015 and 0.00025 in. reflecting a maximum plastic strain at the notch root of 40 (Ti-6Al-4V transverse) and 25 (PH 15-7 Mo) percent. One specimen of Ti-6Al-4V tested in the longitudinal direction showed no measurable change in notch root radius during exposure.

CONCLUSIONS

A summary of all experimental data is presented in Figs. 22 and 23 with the notch strength ratio plotted as a function of the strength to density ratio. This presentation facilitates an easy interpretation of the data with respect to aircraft desirability as expressed by the strength to density ratio and fracture toughness as expressed by the notch strength ratio. Of the materials tested subsequent to exposure the titanium alloy Ti-6Al-4V in the annealed condition seems to be the most promising. Other alloys that warrant further studies at the present time are RS-140 in the annealed condition and RS-115 in the annealed condition. Recent communication with TMCA has indicated that the titanium alloy Ti-8Al-8Mo-1V in the duplex annealed condition might show promising results with respect to the tentative selection criteria indicated. This alloy will be included in the screening program at Syracuse University.

On the basis of poor notch strength values the following alloys investigated to date must be considered unacceptable (these alloys show notch strength ratio of .4 or less in any of the conditions investigated) AISI-301 - 514, PH 15-7 Mo - CH 900, RS-140, solution treated and aged.

P 12

P 13

and

REFERENCES

1. Sell, R. J., Chave, C. and Weiss, V., "Material Evaluation For a Mach III Transport Plane", Report No. 873-617-QP1, NASA (1961).
2. Sell, R. J., Chave, C. and Weiss, V., "Material Evaluation For a Mach III Transport Plane", Report No. 873-611-QP2, NASA (1961).
3. Sell, R. J., Chave, C. and Weiss, V., "Material Evaluation For a Mach III Transport Plane", Report No. 873-622-QP3, NASA (1962).
4. Minutes of First Meeting, Special Committee on Materials Research for Supersonic Transports, NASA, Washington, D. C., May 1961.

TABLE I

MATERIALS

LOY	CONDITION
SI-01	Cold Reduced 31%
SI-301	Cold Reduced 51%
SI-301	Cold Reduced 60%
15-7 Mo	RH 1050 (1750F for 10 minutes, air cooled to 75F in 1 hour, -110F for 22 hours, air warmed to 75F, 1050F for 1 hour, air cooled)
15-7 Mo	CH 900 (Solution treated, aged 1 hour at 900F, air cooled)
6Al-4V (RS-20A)	Annealed
6Al-4V (RS-20A)	Solution treated and aged 4 hours at 925F, air cooled
6Al-3Mo-1V (RS-115)	Annealed
6Al-2.75Cu-1.25Fe (RS-140)	Annealed
6Al-2.75Cu-1.25Fe (RS-141)	Solution treated and aged 200F for 6 hours, air cooled

SUPPLIER	HEAT NUMBER	AMERICAN STEEL & WIRE 390065	REPUBLIC 330522	REPUBLIC 3930269	REPUBLIC 3930265	REPUBLIC 3930017
CARBON	0.06	1.34	0.07	0.034	0.031	0.030
MANGANESE	0.029	0.52	0.015			
PHOSPHORUS	0.013	0.016	0.30			
SULFUR	0.56	7.20	15.34			
SILICON	7.31	17.72				
NICKEL						
CHROMIUM						
MOLYBDENUM						
VANADIUM						
NITROGEN						
ALUMINUM						
HYDROGEN						
IRON						
TITANIUM						

TABLE III
Notch Root Radius Changes During Exposure

MATERIAL	BEFORE EXPOSURE	AFTER EXPOSURE	CHANGE
Ti-6Al-4V (RS-120A) Solution Treated and Aged Spec. No. AVL28B (Long.) Spec. No. AVT323 (Trans.)	0.0005 0.0005	0.0005 0.0007	— 0.0002
Ph 15-7 Mo Condition RH 1050 Spec. No. EV17A (Long.) Spec. No. EVT9A (Trans.)	0.0010 0.0011	0.00125 0.00125	0.00025 0.00015

TABLE IV
Material-AISI-301
34% Cold Reduction

SMOOTH SPECIMENS

Unex-posed	CONDITION		Test Temp °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength-10 ³ in Density = 0.286 lbs in ³
	Exposed 650°F	40 No Load						
X			-110	211.5	-	21.2	24.0	739.5
X			-110	211	155.5	21.0	27.7	737.8
X			75	173.4	166.7	20.4	23.9	606.3
X			75	166.5	-	21.2	-	582.2
X			75	177.8	161	21.2	24.5	621.7
X			350	166.5	151.5	2.4	22.2	582.2
X			650	153	132.5	3.0	21.8	535.0
X			800	134.5	121.5	3.3	19.3	470.3
X			-110	234	186	23.2	27.3	818.2
X			75	204	184	9.0	25.0	13.3
X			650	163.6	149	3.1	25.0	572.0
X		X	75	218	179.5	12.95	26.3	62.2
X			-110	212.5	127	20.1	25.0	743.0
X			75	182.6	143.7	13.9	25.7	638.5
X			350	171	149	3.4	23.4	597.9
X			650	152.5	-	3.3	-	533.2
X			800	142	126.5	2.6	21.8	544.2
X			-110	210.1	117	3.0	22.8	496.5
X			75	201	187.7	9.7	29.3	734.6
X			650	169.2	171	1.6	27.0	702.8
X			-110	216	140	3.1	25.0	591.6
X		X	75	195	179.5	9.39	25.65	555.2
X		X			177	8.68	30.2	681.8

TABLE V
Material-AISI-301
34% Cold Reduction
NOTCH SPECIMENS

Rolling Direction	Unex-posed	CONDITION		Notch <0.001	Test Temp °F	Notch Strength KSI	Notch Strength Ratio
		Exposed 650°F	40 No Load				
L	X			X	-110	183	0.865
L	X			X	75	187.4	1.087
L	X			X	350	172.6	1.036
L	X			X	650	137.5	0.899
L	X			X	800	117.5	0.884
L	X			X	75	194.5	1.126
L	X			X	-110	250	0.940
L	X			X	75	209	1.023
L	X			X	650	126	0.771
L	X			X	-110	210	1.149
L	X			X	75	206	0.945
T	X			X	-110	164.5	0.774
T	X			X	-110	164	0.772
T	X			X	75	167.5	0.917
T	X			X	75	164	0.898
T	X			X	350	146	0.854
T	X			X	650	130	0.760
T	X			X	800	102.9	0.662
T	X			X	650	110.5	0.710
T	X			X	800	138	0.761
T	X			X	800	98	0.890
T	X			X	75	172	0.942
T	X			X	-110	187.4	0.892
T	X			X	-110	193	0.904
T	X			X	75	190	0.945
T	X			X	75	187	0.930
T	X			X	650	96.2	0.569
T	X			X	650	102	0.603
T	X			X	-110	178	0.824
T	X			X	75	188	0.964

TABLE VI
Material-AISI-301
51% Cold Reduction

SMOOTH SPECIMENS

Direction	Unex-posed	CONDITION		Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elasticity E-10 ⁶ PSI	Strength Density = $\frac{\text{Strength}}{0.286 \text{ in}^3}$ 10 ³ in ³
		Exposed 650°F 1000 Hrs.	Load					
L	X			223.5	215	18.0	22.9	781.5
L	X			216	211.5	2.3	23.4	755.2
L	X			350	188.5	2.0	24.5	727.3
L	X			650	191	2.6	25.1	667.8
L	X			800	165	2.6	19.7	576.9
L	X			110	253.5	6.35	24.8	909.1
L	X			260	294	1.47	27.35	909.1
L	X			75	219	1.67	24.3	765.1
L	X			650	201.5	10.0	29.6	979.0
L	X			110	245.5	2.34	27.4	926.4
L	X			75	265			
T	X			238	196.5	11.4	26.1	832.2
T	X			214	194	3.6	26.3	748.3
T	X			350	187.5	2.4	25.7	736.0
T	X			650	178	2.4	24.5	713.3
T	X			800	145.2	2.3	22.3	604.9
T	X			110	251	5.68	32.3	961.5
T	X			75	246.5	-	28.1	910.1
T	X			650	208.5	1.67	25.4	866.7
T	X			110	278	6.35	30.5	972.0
T	X			75	267	1.67	31.4	933.4

TABLE VII

Material-AISI-301
51% Cold Reduction

NOTCH SPECIMENS

Direction	Unex-posed	CONDITION		Notch <0.01	Natural Crack	Test Temp °F	Notch Strength KSI	Notch Strength Ratio
		Exposed 650°F 1000 Hrs.	Load					
L	X			X		-110	237	1.060
L	X			X		75	214.5	0.993
L	X			X		350	152	0.731
L	X			X		650	123	0.644
L	X			X		800	121.1	0.734
L	X			X		110	207	0.958
L	X			X		75	237	0.912
L	X			X		350	221	0.850
L	X			X		650	103	0.470
L	X			X		110	240	0.857
L	X			X		75	229	0.864
T	X			X		-110	176	0.739
T	X			X		75	171	0.718
T	X			X		350	163	0.762
T	X			X		650	153	0.715
T	X			X		800	128.5	0.610
T	X			X		110	129	0.613
T	X			X		75	103	0.505
T	X			X		350	95.6	0.469
T	X			X		650	104.2	0.602
T	X			X		800	104	0.601
T	X			X		110	165.1	0.771
T	X			X		75	153	0.556
T	X			X		-110	147.5	0.536
T	X			X		75	146	0.549
T	X			X		350	170.5	0.641
T	X			X		650	75.7	0.336
T	X			X		800	78.2	0.348
T	X			X		110	150	0.540
T	X			X		75	165.5	0.620

TABLE VII
Material-AISI-301
60% Cold Reduction

Unex-posed	CONDITION		Test Temp. °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength Density = $\frac{103 \text{ in}}{0.286 \text{ lbs}} \left(\frac{\text{in}^3}{\text{in}^3} \right)$
	Exposed 1000 Hrs.	Exposed 40 KSI Load						
X	X		-110	231	206	10.7	25.1	807.7
X	X		75	223.5	222.5	2.4	25.7	781.5
X	X		350	213.5	207.5	2.4	22.5	746.5
X	X		650	194.5	180	2.6	22.9	680.1
X	X		800	174.5	166.7	3.3	20.9	610.1
	X	X	-110	278	276.3	3.0	26.3	972.0
	X	X	75	269	272.5	1.67	25.65	940.6
	X	X	650	216	-	1.33	-	755.2
		X	-110	284	276	4.68	27.15	993.0
		X	75	258	251	2.01	27.9	902.1
X	X		-110	240	205.5	4.5	27.9	839.2
X	X		75	222	210	2.7	29.5	776.2
X	X		350	225	203.5	2.0	29.5	766.7
X	X		650	190	175.5	1.4	25.8	664.3
X	X		800	208.5	181	3.0	25.8	729.0
X	X		800	183.8	160	3.0	25.0	642.7
X	X		800	183.8	-	3.0	-	642.7
	X	X	-110	290	280	1.34	30.15	1014.0
	X	X	75	269	259.5	1.3	30.15	940.6
	X	X	650	204	-	0.99	-	713.3
		X	-110	288	270	4.0	32.2	1007.0
		X	75	261	264	1.67	31.25	982.5

TABLE IX
Material-AISI-301
50% Cold Reduction
NOTCH SPECIMENS

Unex-posed	C	NOTCH		Notch Depth <0.001	Natural Fracture	Test Temp. °F	Notch Strength KSI	Notch Strength Ratio
		Ex	ON 0°F					
X	X					-110	226.5	0.981
X	X					75	206.5	0.924
X	X					350	159	0.745
X	X					650	128.5	0.661
X	X					800	135	0.774
X	X				X	-110	232	1.004
X	X				X	-110	231.5	1.002
X	X				X	75	206.2	0.923
		X				-110	248.5	0.894
		X				75	218	0.810
		X				650	122	0.565
		X				-110	260	0.915
		X				75	235	0.911
X	X					-110	184.5	0.769
X	X					75	173.5	0.723
X	X					350	167.2	0.753
X	X					650	142.2	0.668
X	X					800	121.5	0.540
X	X					800	114	0.507
X	X					650	136.2	0.534
X	X					650	106	0.533
X	X					800	82.4	0.339
X	X					800	93.2	0.507
X	X					800	139.5	0.596
X	X				X	-110	167.5	0.698
X	X				X	-110	167	0.696
X	X				X	75	159.5	0.718
		X				-110	135	0.466
		X				-110	134.2	0.463
		X				75	153	0.569
		X				75	132	0.491
		X				650	97.5	0.478
		X				650	90.5	0.444
		X				-110	177	0.615

TABLE I

Material-PH 15-7 Mo
Heat Treatment RH 1050

SMOOTH SPECIMENS

Direction	CONDITION			Test Temp. °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength 10 ³ in Density (Density = 0.286 lbs/in ³)
	Unexposed	Exposed 650°F 1000 Hrs	No Load						
L	X			-110	218	212	5.47	30.2	78.0
L	X			75	201	196.5	4.34	27.1	72.6
L	X			350	184	180	3.01	24.5	66.3
L	X			650	169	159.5	3.34	25.7	61.0
L	X			800	143.5	139	2.01	21.0	518.1
L		X		-110	232	226.5	6.35	32.1	837.5
L		X		75	214	211	4.01	27.8	772.6
L		X		650	178	167	4.01	29.35	642.6
L			X	-110	232	227	4.68	28.6	837.5
L			X	75	215	211	4.01	26.4	776.2
T				-110	224.5	219	5.95	29.35	810.5
T	X			75	204	197.5	3.68	28.55	736.5
T	X			350	186.5	182.5	2.34	27.75	673.3
T	X			650	170	161	2.68	33.1	613.7
T	X			800	152	141	5.35	24.5	548.7
T		X		-110	230	224.5	3.68	29.4	830.3
T		X		75	214	209	3.01	30.2	772.5
T		X		650	176.8	172	2.68	29.35	638.3
T		X	X	-110	234	225	4.68	31.2	844.8
T			X	75	217	213	2.68	29.4	783.4

TABLE II

Material-PH 15-7 Mo
Heat Treatment RH 1050

NOTCH SPECIMENS

Rolling Direction	Unexposed	CONDITION		Notch <0.001	Natural Crack	Test Temp. °F	Notch Strength KSI	Notch Strength Ratio
		Exposed 650°F 1000 Hrs	No Load					
L	X			X		-110	139	0.638
L	X			X		75	155.2	0.921
L	X			X		350	155	0.842
L	X			X		650	154	0.911
L	X			X		800	150	1.045
L	X				X	-110	160	0.826
L	X				X	-110	121.5	0.557
L	X				X	75	196.5	0.976
L	X			X		-110	195.2	0.453
L	X			X		75	173	0.808
L	X			X		650	167.5	0.941
L	X		X	X		-110	138.5	0.596
L	X		X	X		75	178	0.827
T	X			X		-110	116	0.517
T	X			X		-110	87	0.388
T	X			X		75	152	0.34
T	X			X		75	144	0.745
T	X			X		350	143	0.706
T	X			X		650	149	0.768
T	X			X		800	140	0.799
T	X			X		-110	110	0.647
T	X			X		650	127	0.706
T	X			X		800	137	0.901
T	X			X		-110	134	0.882
T	X			X		-110	144.5	0.644
T	X			X		75	145	0.846
T	X			X		75	169.5	0.831
T	X	X		X		-110	98	0.436
T	X	X		X		-110	100	0.434
T	X	X		X		75	122	0.570
T	X	X		X		75	118	0.551
T	X	X		X		650	116.5	0.658
T	X	X		X		650	111.3	0.629
T	X		X	X		-110	92.5	0.395
T	X		X	X		75	144.4	0.665

TABLE XII

Material-PH 15-7 Mo
Heat Treatment CH 900

SMOOTH SPECIMENS

Unex- posed	CONDITION			Test Temp. °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength-10 ³ in Density (Density = 0.286 lbs in ³)
	Exposed 650°F								
	40 KSI	No Load	1000 Hrs.						
X				-110	304	305	2.0	28.9	1097.4
X				75	283.9	279	1.0	28.9	1024.9
X				350	246	-	1.34	-	895.3
X				650	224	217.5	1.34	25.9	838.6
X				800	201	-	2.34	-	725.6
X				-110	307	294	3.67	32.4	1108.3
X				75	292	278	0.67	32.4	1054.1
X				350	257	230	1.34	-	927.7
X				650	236	206	2.34	33.4	851.9
X				800	220	205.5	2.0	30.6	794.2

TABLE XIII

Material-PH 15-7 Mo
Heat Treatment CH900

NOTCH SPECIMENS

Rolling Direction	Unex- posed	CONDITION			Notch <0.001	Crack Natural	Test Temp °F	Notch Strength KSI	Notch Strength Ratio
		Exposed 650°F							
		40 KSI	No Load	1000 Hrs					
L	X				X	-110	236	0.776	
L	X				X	75	229	0.807	
L	X				X	75	225	0.193	
L	X				X	350	177	0.714	
L	X				X	650	166	0.741	
L	X				X	800	156	0.776	
T	X				X	-110	126.3	0.411	
T	X				X	-110	127.5	0.415	
T	X				X	75	126	0.438	
T	X				X	75	128	0.438	
T	X				X	350	114	0.444	
T	X				X	350	100.5	0.422	
T	X				X	650	96	0.407	
T	X				X	650	92.5	0.337	
T	X				X	800	95.5	0.354	
T	X				X	800	82.4	0.344	

TABLE IV

Material-Ti-6Al-4V (RS-120A)
Heat Treatment Annealed

SMOOTH SPECIMENS

Unex- posed	CONDITION			Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength-10 ³ in Density (Density = 0.160 lbs in ³)
	Exposed 25 KSI	100 Hrs. No	Load					
X				-110 170.5	163.5	6.35	20.1	1065.6
X				75 135	124.5	9.7	15.8	843.7
X				350 115.5	102.5	11.36	16.55	721.9
X				650 131	75	8.7	14.2	631.3
X				800 91.4	75.4	13.03	13.5	571.3
				-110 173	163	11.3	16.55	1081.3
		X		75 146.8	135	10.0	17.1	917.5
		X		650 100.5	80.5	9.05	16.8	628.1
				-110 169.5	159	12.6	17.7	1059.4
X				75 146	136	13.0	18.3	912.5
X				350 123.5	110.5	10.38	17.4	711.9
X				650 107.8	89.7	6.35	15.1	673.7
X				800 100	83.6	5.35	15.3	625.0
		X		-110 175	166.3	11.0	19.35	1093.7
		X		75 152	140	12.5	14.05	950.0
		X		650 111.2	93.6	7.03	13.85	695.0

TABLE XV

Material-Ti-6Al-4V (RS-120A)
Heat Treatment Annealed

NOTCH SPECIMENS

Direction Rolling	Unex- posed	CONDITION			Notch <0.001	Natural Crack	Test Temp °F	Notch Strength KSI	Notch Strength Ratio
		Exposed 25 KSI	1000 Hrs No	Load					
L	X				X		-110	136.6	0.801
L	X				X		75	137.2	1.016
L	X				X		350	118	1.022
L	X				X		650	99	0.980
L	X				X		800	94.5	1.034
L	X					X	-110	137	0.804
L	X					X	75	123	0.721
L	X					X	-110	118	0.874
L			X		X		75	134.5	0.777
L			X		X		75	137.6	0.937
L			X		X		650	102.5	1.020
T	X				X		-110	131.2	0.774
T	X				X		-110	134.6	0.794
T	X				X		75	137.2	0.940
T	X				X		75	142.9	0.979
T	X				X		350	132	1.069
T	X				X		350	127	1.028
T	X				X		650	109.6	1.017
T	X				X		800	107.5	0.997
T	X				X		800	110	1.100
T	X				X		800	105.8	1.058
T	X				X	X	-110	106	0.625
T	X				X	X	-110	133	0.785
T	X				X	X	75	129	0.884
T			X		X		-110	142.8	0.816
T			X		X		-110	140.5	0.803
T			X		X		75	143	0.941
T			X		X		75	139.5	0.918
T			X		X		650	136	0.953
T			X		X		650	112.5	1.012

TABLE XVI

Material-Ti-6Al-4V (RS-120A)
Heat Treatment Solution Treated,
Aged 4 Hrs. 935F

SMOOTH SPECIMENS

Unex- posed	CONDITION			Test Temp. °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength-10 ³ in Density (Density = 0.160 lbs / in ³)
	Exposed 650°F								
	40 KSI	1000 Hrs.	Load						
X				-110	197.5	182.5	7.03	15.8	1234.4
X				75	171	155	6.01	18.0	1068.7
X				350	145.8	118.5	7.7	15.55	911.3
X				650	126.3	97.4	5.35	15.3	789.4
X				800	121	92.9	5.67	16.3	756.3
				-110	199	165.5	2.68	17.7	1243.7
	X			75	171.5	-	5.02	-	1071.9
	X			650	131.3	101	5.01	14.25	820.6
	X			75	165	151.5	6.02	15.6	1031.3
		X	X	75	162	152.5	6.02	15.3	1012.5
				-110	208	194.5	7.0	17.7	1300.0
X				75	164.2	151.5	8.36	17.7	1026.3
X				350	143.8	120	6.02	15.1	698.7
X				650	135.2	104.7	5.30	17.4	645.0
X				800	127.3	95.9	5.01	18.3	795.5
				-110	206	190	2.0	17.7	1287.5
	X			75	165	162	0.67	18.3	1031.3
	X			650	133.2	105	4.68	13.5	832.5
		X	X	-110	197	182.5	4.68	17.1	1231.3
		X		75	2.9	-	6.35	-	1116.7

TABLE XVII

Material-Ti-6Al-4V (RS-120A)
Heat Treatment Solution Treated, aged 4 Hrs. 935F

NOTCH SPECIMENS

Rolling Direction	Unex- posed	CONDITION			Notch Crack Natural	Test Temp °F	Notch Strength KSI	Notch Strength Ratio
		Exposed 650°F						
		25 KSI	1000 Hrs. No	Load				
L	X				X	-110	111.5	0.565
L	X				X	75	130	0.760
L	X				X	350	135	0.926
L	X				X	650	117	0.926
L	X				X	800	103.8	0.858
L	X				X	-110	116.5	0.590
L	X				X	-110	111.5	0.564
L	X				X	75	122	0.714
L	X				X	-110	112.5	0.565
L	X	X	X	X	X	75	119.5	0.697
L	X	X	X	X	X	650	126.8	0.966
L					X	-110	103.5	0.498
T	X				X	-110	100.5	0.483
T	X				X	75	127	0.773
T	X				X	75	119.1	0.725
T	X				X	350	141	0.981
T	X				X	650	142	0.987
T	X				X	650	124	0.917
T	X				X	650	123	0.910
T	X				X	800	116.4	0.914
T	X				X	800	115.5	0.907
T	X				X	-110	90.5	0.435
T	X				X	-110	93	0.447
T	X				X	75	120	0.731
T	X				X	-110	100.5	0.488
T					X	-110	104	0.505
T					X	75	125	0.758
T					X	75	116.5	0.706
T					X	650	124.8	0.937
T					X	650	130	0.976
T					X	-110	103.8	0.527
T				X	X	75	111	0.620

TABLE XVIII

Material-Ti-6Al-3Mo-1V (RS-115)
Heat Treatment Annealed

SMOOTH SPECIMENS

CONDITION			Test Temp. °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength-10 ³ in ³ Density = (0.163 lbs in ³)
Unex- posed	Exposed 650°F 1000 Hrs							
	No Load	25 KSI						
X			-110	164	154.5	5.02	16.4	1006.1
X			75	134.5	127	8.7	15.05	825.2
X			350	115.5	97.7	8.7	14.6	708.6
X			650	95.5	76.2	4.69	14.05	585.9
X			800	86.5	73.1	4.35	12.25	530.7
X			-110	164.5	158	4.10	16.2	1009.2
X			75	135	128	8.38	15.6	828.2
X			350	114	105	6.02	17.2	599.4
X			650	93.5	79.4	4.69	11.85	573.0
X			800	88	72	2.66	16.2	549.9

TABLE XIX

Material-Ti-6Al-3Mo-1V (RS-115)
Heat Treatment Annealed

NOTCH SPECIMENS

Rolling Direction	Unex- posed	CONDITION			Notch Crack Natural	Test Temp °F	Notch Stress KSI	Notch Strength Ratio
		Exposed 650°F	1000 Hrs					
			25 KSI	No				
				Load				
L	X					-110	124	0.756
L	X					75	128	0.951
L	X					350	125	1.082
L	X					650	104	1.089
L	X					800	97	1.121
L	X				X	-110	104	0.634
L	X				X	-110	121	0.737
L	X				X	75	115.6	0.859
T	X					-110	101	0.613
T	X				X	-110	124	0.753
T	X				X	75	132	0.977
T	X				X	75	138	1.022
T	X				X	350	118.5	1.039
T	X				X	350	119	1.043
T	X				X	650	98.5	1.053
T	X				X	650	104	1.112
T	X				X	800	97	1.102
T	X				X	800	98	1.113
T	X				X	-110	113.6	0.686
T	X				X	-110	104.6	0.632
T	X				X	75	127	0.940

TABLE XI

Material-Ti-5Al-2.75Cr-1.25 Fe (RS-140)
Heat Treatment Annealed

SMOOTH SPECIMENS

Unex- posed	CONDITION			Test Temp. °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E-10 ⁶ PSI	Strength-10 ³ lb/in ³ (Density = 0.163 lbs in ³)
	Exposed 650°F								
	25 KSI	1000 Hrs.	No Load						
X				-110	169	165.5	6.0	16.85	1036.8
X				75	146	132	13.6	17.15	895.7
X				350	116.3	100.8	9.38	15.55	725.8
X				650	109	85.3	7.68	15.35	668.7
X				800	95.2	79.7	12.85	12.4	584.0
X				-110	164	153	12.7	16.6	1006.1
X				75	143	132.5	13.05	15.8	877.3
X				350	120.5	99.7	11.7	14.5	739.3
X				650	106	85.9	8.7	15.6	650.3
X				800	95.6	73.5	13.18	15.6	586.5
	X			75	147	133.5	11.5	17.1	901.8

TABLE XII

Material-Ti-5Al-2.75Cr-1.25Fe (RS-140)
Heat Treatment Annealed

NOTCH SPECIMENS

Rolling Direction	Unex-posed	CONDITION			Notch KSI	Crack	Test Temp °F	Notch Strength KSI	Notch Strength Ratio
		Exposed 650°F							
		25	1000 Hrs.	No Load					
L	X				X		-110	136	0.804
	X				X		75	136	0.931
	X				X		350	127.4	1.076
	X				X		650	113.5	1.041
	X				X		800	100	1.050
	X				X		-110	125.5	0.742
	X				X		-110	115	0.680
	X				X		75	129	0.883
	X				X		75	131	
	X				X		650	113	
T	X				X		-110	124	0.756
	X				X		-110	132.2	0.806
	X				X		75	146	1.020
	X				X		75	144.5	1.010
	X				X		350	122	1.012
	X				X		350	126.5	1.062
	X				X		650	117.6	1.109
	X				X		650	118.2	1.115
	X				X		800	112.5	1.176
	X				X		800	114	1.192
T	X				X		-110	106	0.646
	X				X		-110	116.4	0.709
	X				X		75	145	1.013
	X				X		75	144.8	0.985
	X				X		75	154	1.046
	X				X		650	109	
	X				X		650	119	
	X				X				
	X				X				
	X				X				

TABLE XIII

Material-Ti-5Al-2.75Cr-1.25Fe (RS-140)
Heat Treatment Solution Treated, Aged 900F 6hrs.

SMOOTH SPECIMENS

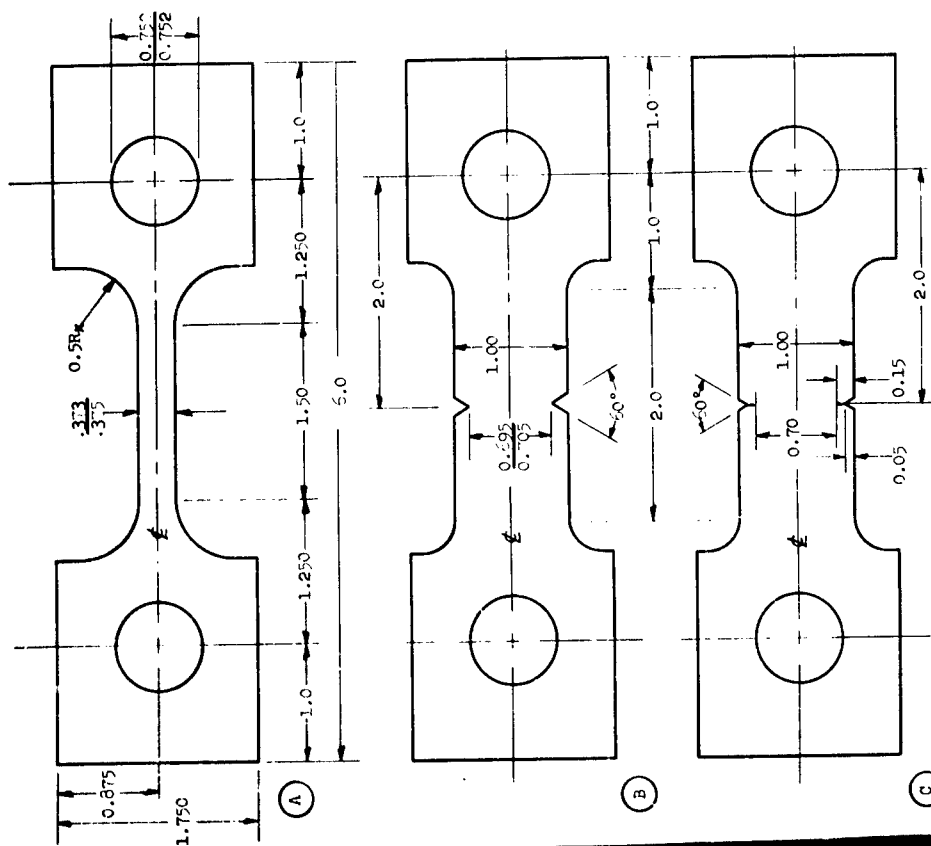
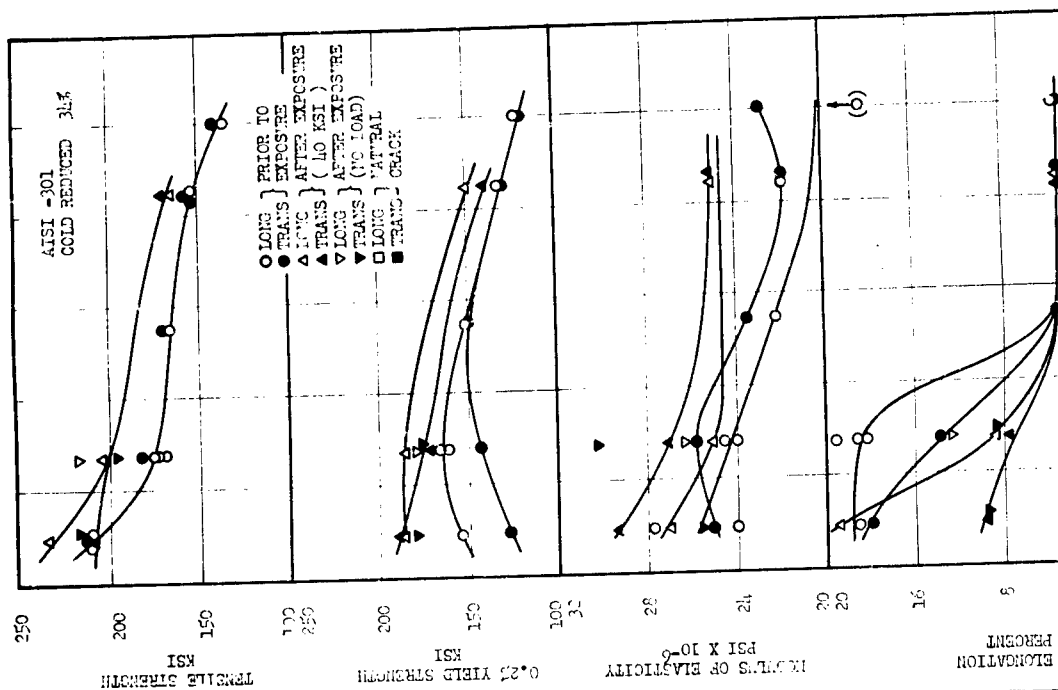
Unex- posed	CONDITION			Test Temp. °F	Tensile Strength KSI	0.2% Yield Strength KSI	Elong. Percent (in 1.5")	Mod. of Elas. E=10 ⁶ PSI	Strength-10 ³ in Density (Density = 0.163 lbs) in ³
	Exposed	Exposed 650°F							
		25 KSI	No Lo-d						
X				-110	206	179.5	1.67	16.3	1263.8
X				75	171	143.5	4.35	15.0	1049.0
X				350	158	118.5	6.22	17.75	969.3
X				350	151	115.5	5.0	14.15	926.3
X				650	148.5	102	6.0	16.9	911.0
X				800	110.5	83	9.03	12.6	677.9
		X		-110	205	170.5	4.01	16.9	1257.6
		X		75	177.5	146.5	4.68	16.55	1088.9
	X			-110	196	183	2.01	19.1	1202.4
	X			75	172.8	145.5	3.01	15.4	1060.1
	X			650	145	99	5.35	14.15	689.5
				-110	206	181	3.34	17.4	1263.8
X				75	176.5	153	5.35	16.9	1082.6
X				350	150	118.5	6.22	20.6	920.2
X				650	134	94.2	6.48	20.6	822.0
X				800	113.6	86.1	9.38	11.0	696.9
		X		-110	213	182	3.01	17.5	1306.7
		X		75	184	154	6.36	16.9	1126.8
			X	-110	202	180.5	2.34	17.3	1239.3
		X		75	176.5	153.5	2.34	17.2	1082.8
		X		650	142	100	5.35	15.4	871.2

TABLE XIII

Material-Ti-5Al-2.75Cr-1.25Fe (RS-140)
Heat Treatment Solution Treated, Aged 900F 6hrs.

NOTCH SPECIMENS

Rolling Direction	Unexposed	CONDITION			Notch Crack Natural	Test Temp °F	Notch Strength KSI	Notch Strength Ratio
		Exposed	650°F					
			25 KSI	No Load				
L	X				-110	90	0.437	
L	X				75	93.4	0.546	
L	X				350	116.5	0.737	
L	X				650	131	0.882	
L	X				800	129.5	1.172	
L				X	-110	39	0.190	
L				X	75	58	0.326	
L			X		-110	42.3	0.216	
L			X		75	65.4	0.378	
L			X		650	112.5	0.776	
T	X				-110	90	0.437	
T	X				-110	87.5	0.425	
T	X				75	93.4	0.529	
T	X				75	94.5	0.535	
T	X				350	114	0.760	
T	X				350	115	0.767	
T	X				650	131.5	0.961	
T	X				650	118	0.880	
T	X				800	131.5	1.157	
T	X				800	123.5	1.131	
T				X	-110	52	0.244	
T				X	75	61.3	0.333	
T				X	-110	49	0.243	
T			X		-110	61	0.302	
T			X		75	79.9	0.453	
T			X		75	77.8	0.441	
T			X		650	122.5	0.863	
T			X		650	125.6	0.885	



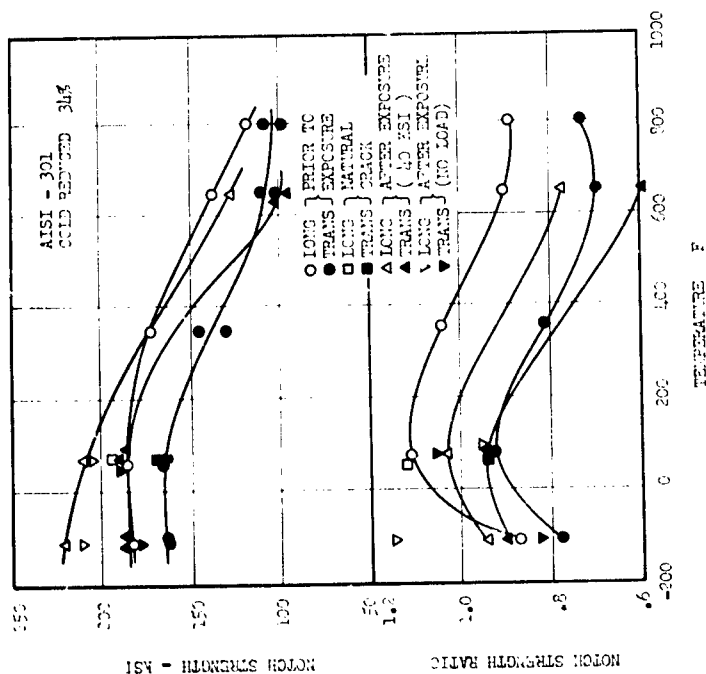
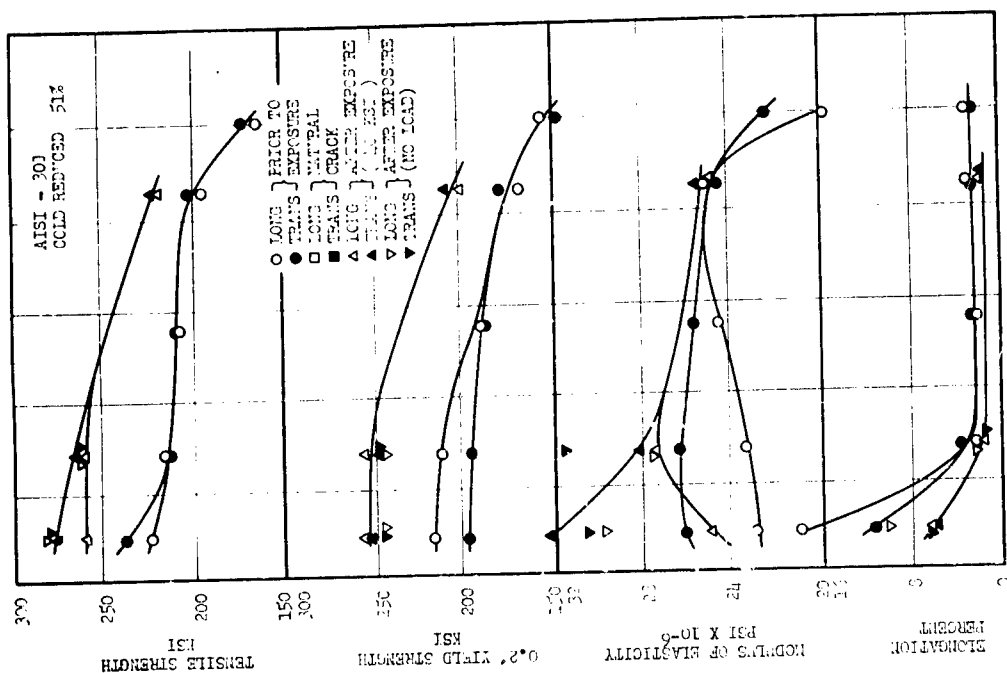
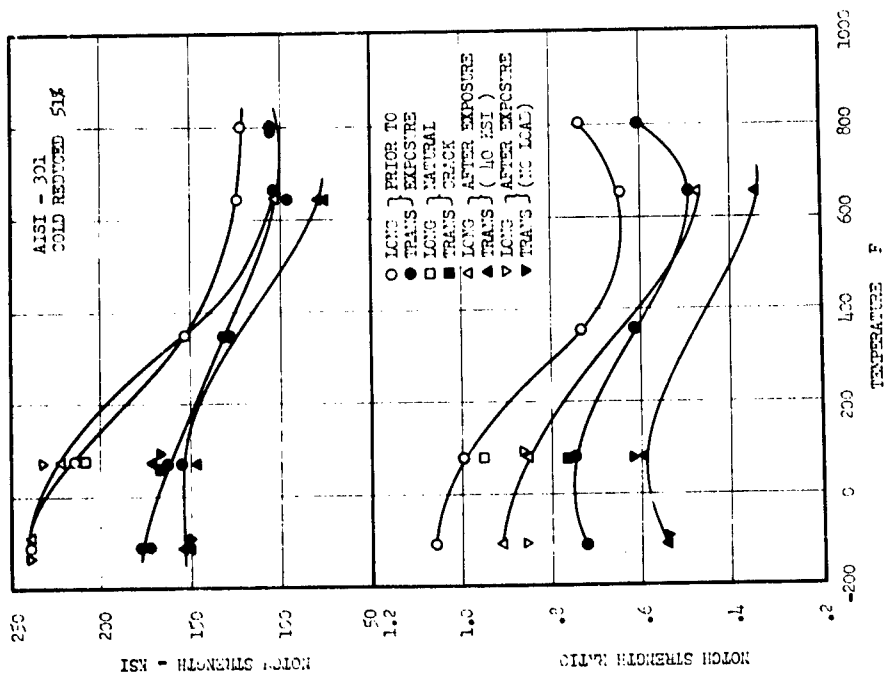
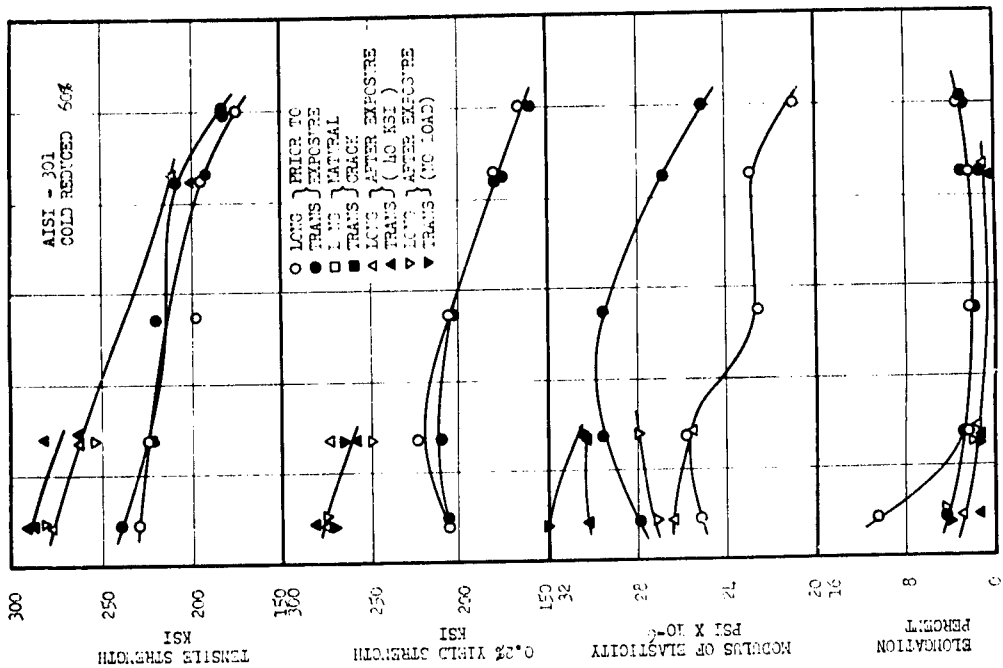


FIG. 3 THE EFFECT OF TESTING TEMPERATURES ON THE NOTCH STRENGTH AND NOTCH STRENGTH RATIO OF AISI - 301 COLD REDUCED 34% MATERIAL TESTED PRIOR TO EXPOSURE AND IN THE EXPOSED CONDITION.



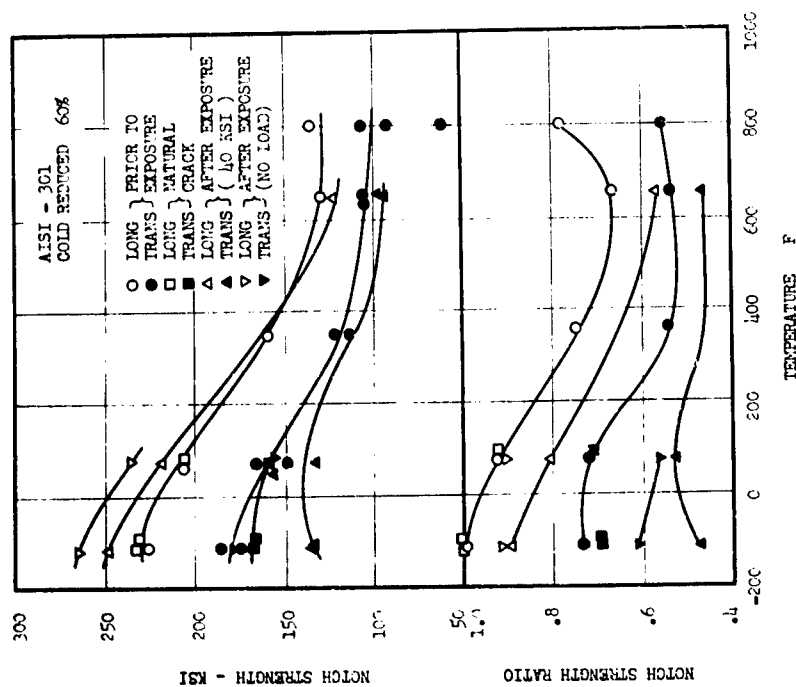
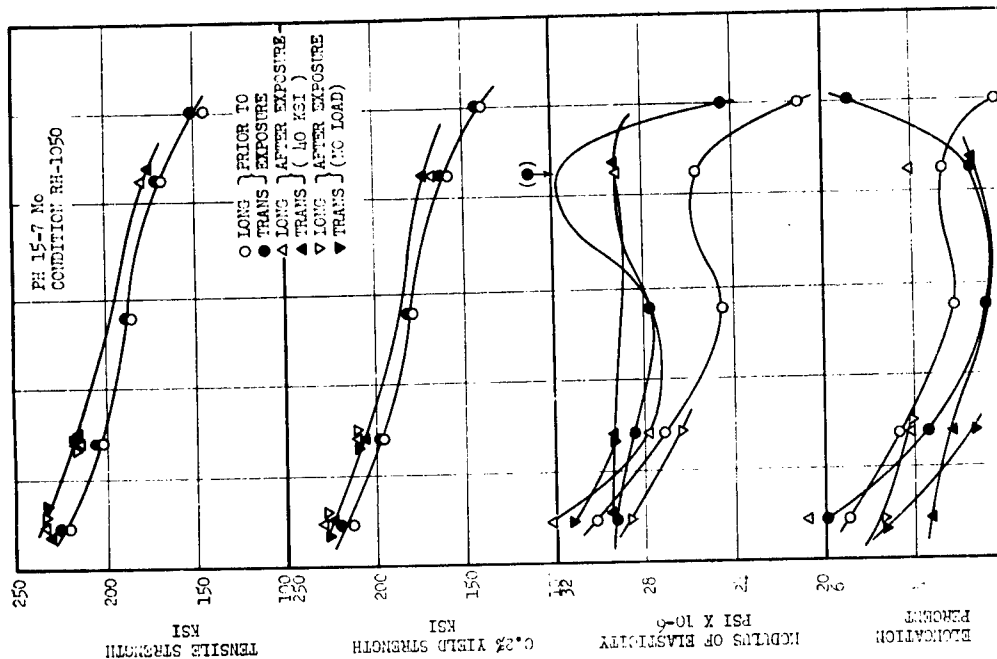
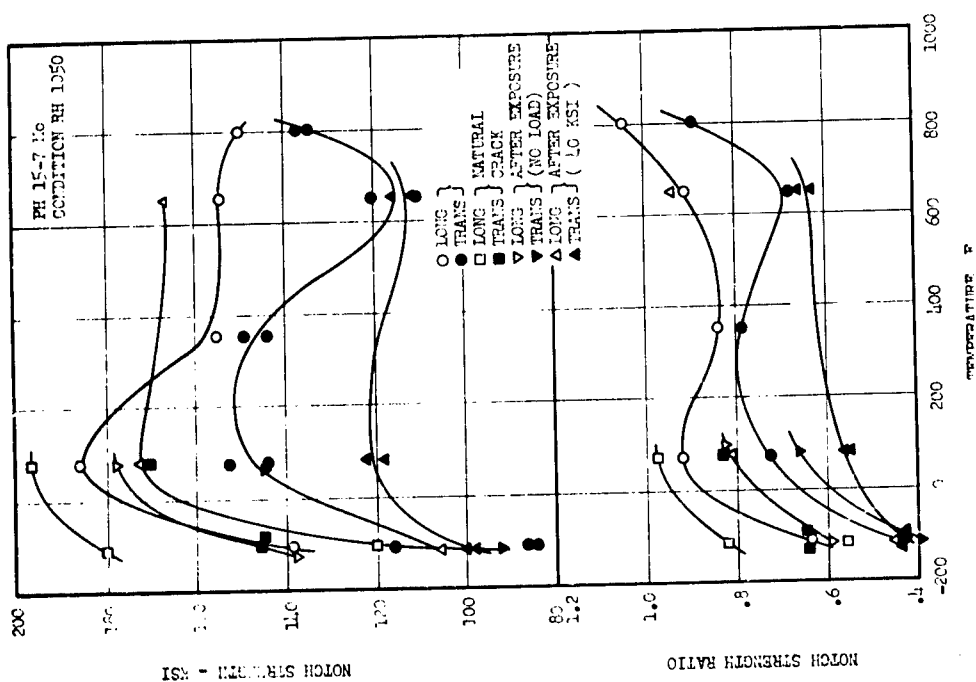
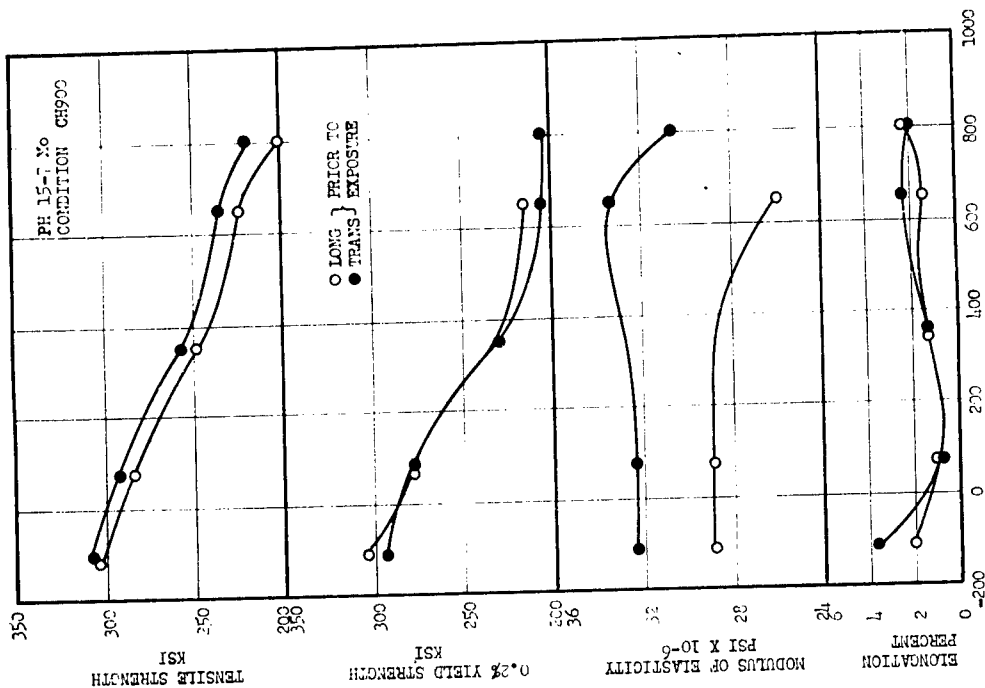
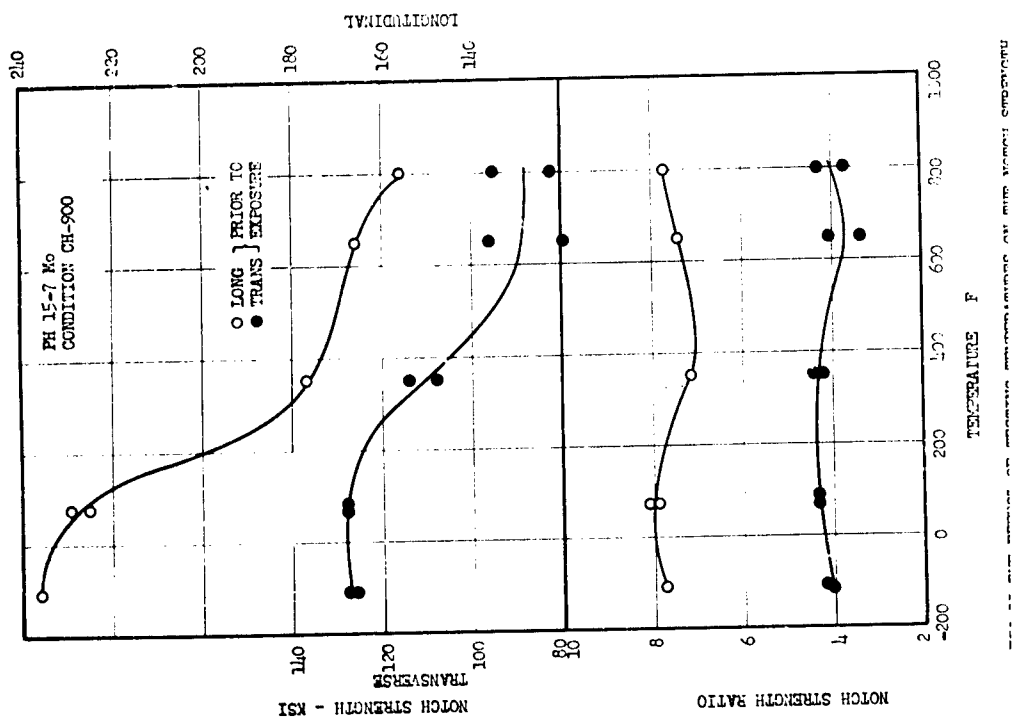
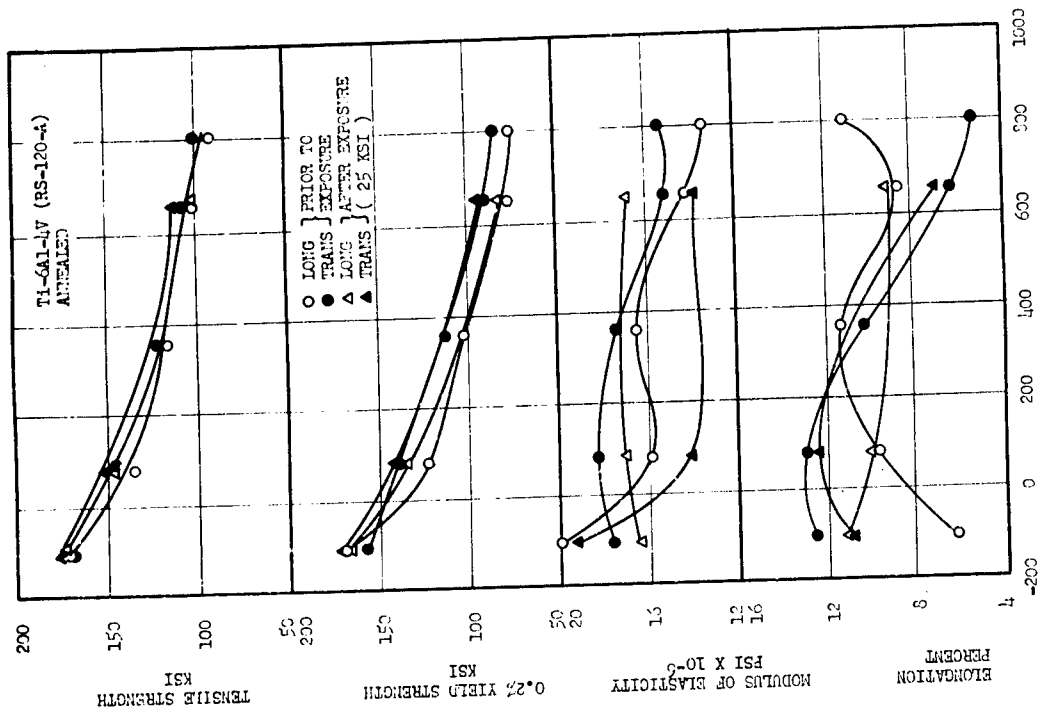


FIG. 7 THE EFFECT OF TESTING TEMPERATURES ON THE NOTCH STRENGTH AND NOTCH STRENGTH RATIO OF AISI - 301 COLD REDUCED 60%.





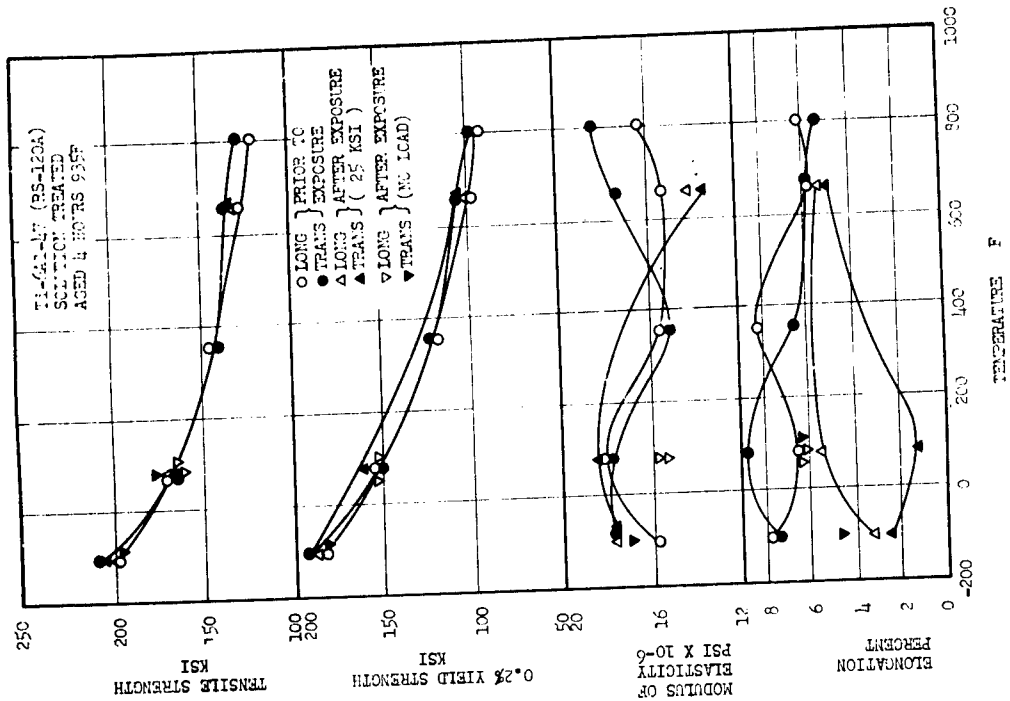


FIG. 13 THE EFFECT OF TESTING TEMPERATURES ON THE NOTCH STRENGTH AND NOTCH STRENGTH RATIO OF Ti-6Al-4V (RS-120-A) ANNEALED, MATERIAL TESTED PRIOR TO EXPOSURE AND IN THE EXPOSED CONDITION.

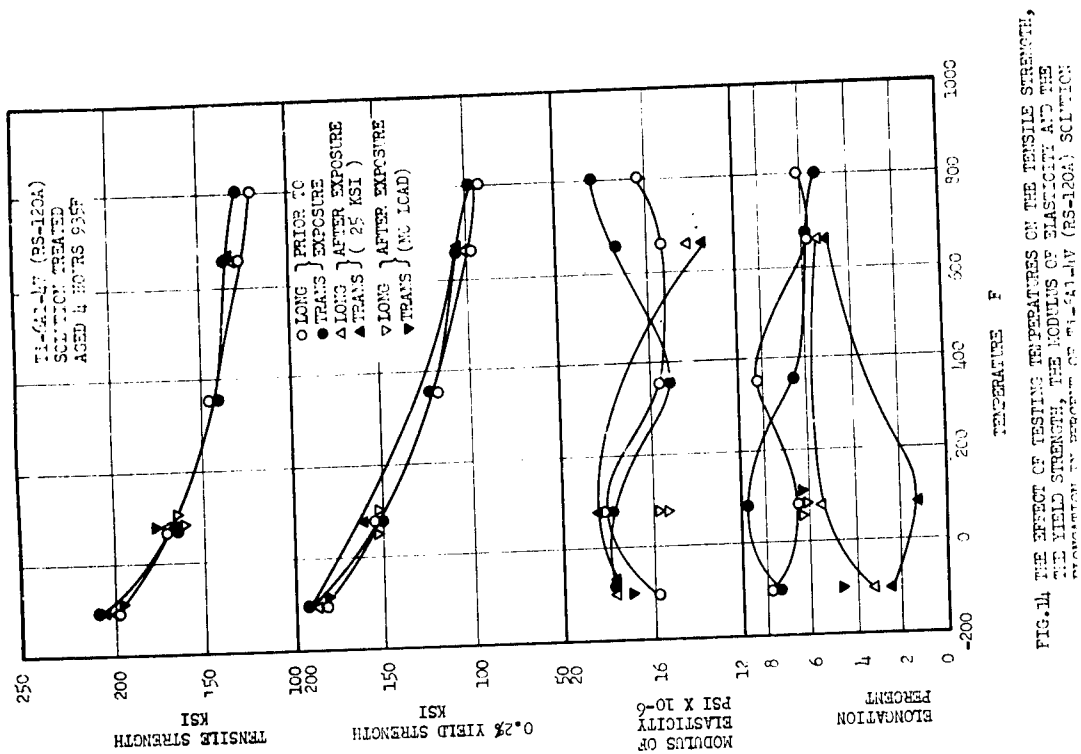


FIG. 14 THE EFFECT OF TESTING TEMPERATURES ON THE TENSILE STRENGTH, THE YIELD STRENGTH, THE MODULUS OF ELASTICITY AND THE ELONGATION OF Ti-6Al-4V (RS-120-A) SOLUTION TREATED, MATERIAL TESTED PRIOR TO EXPOSURE AND IN THE EXPOSED CONDITION.

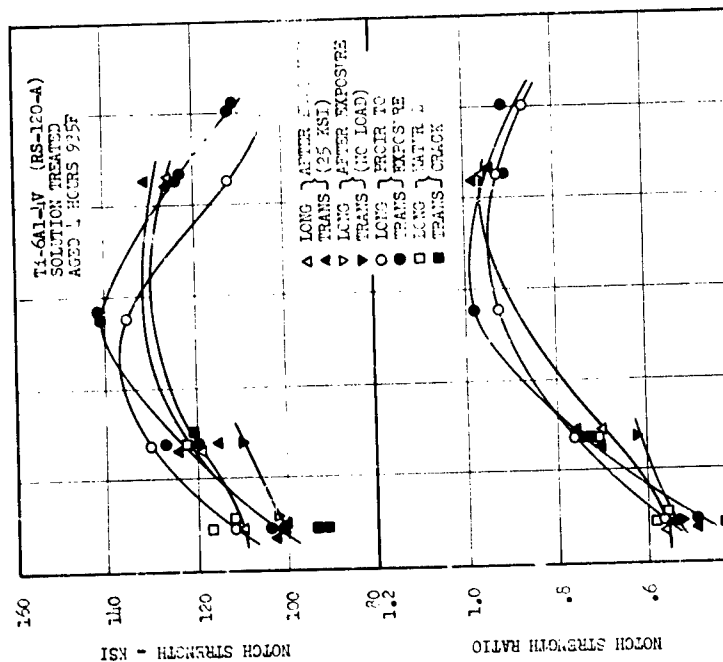
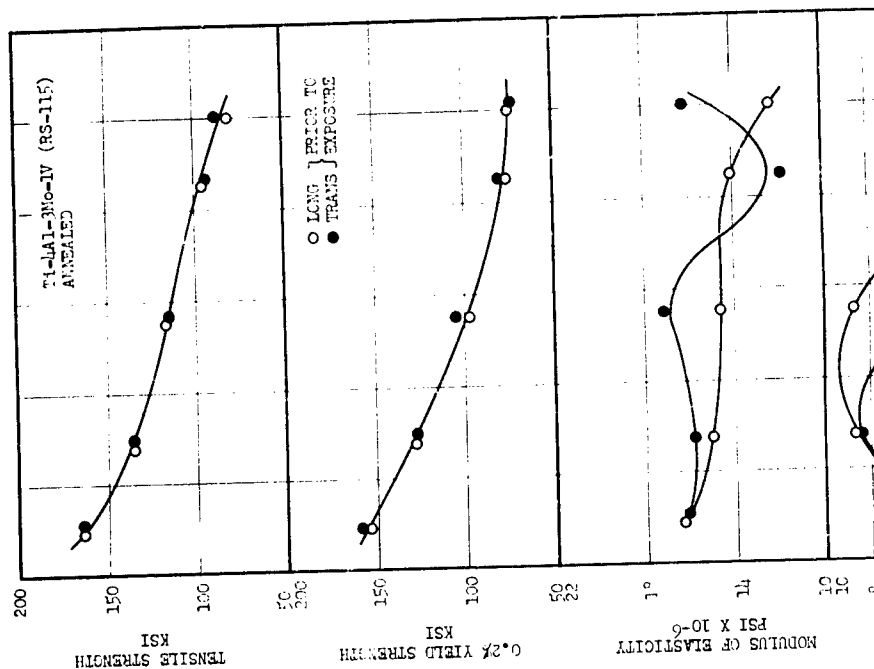


FIG. 4 THE EFFECT OF TESTING TEMPERATURES ON THE TENSILE STRENGTH, ELONGATION, AND MODULUS OF ELASTICITY AND THE YIELD STRENGTH, OF AISI - 301 COLD REDUCED 91% ALUMINUM IN PRESENT OF AISI - 301 COLD REDUCED 91% ALUMINUM TESTED PRIOR TO EXPOSURE AND IN THE EXPOSED CONDITION.



CONTINUED.

42

ERRATA SHEET
FINAL REPORT - JUNE 62
NASA
CONTRACT NO. NASr-43

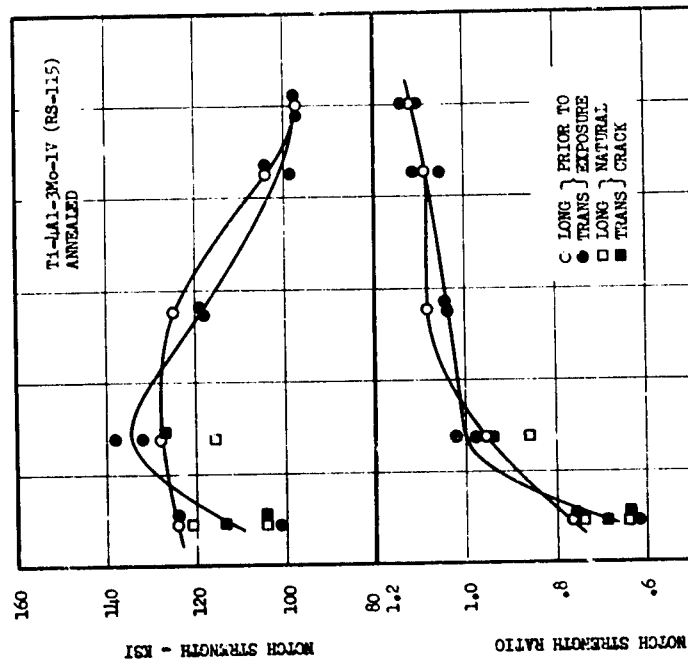
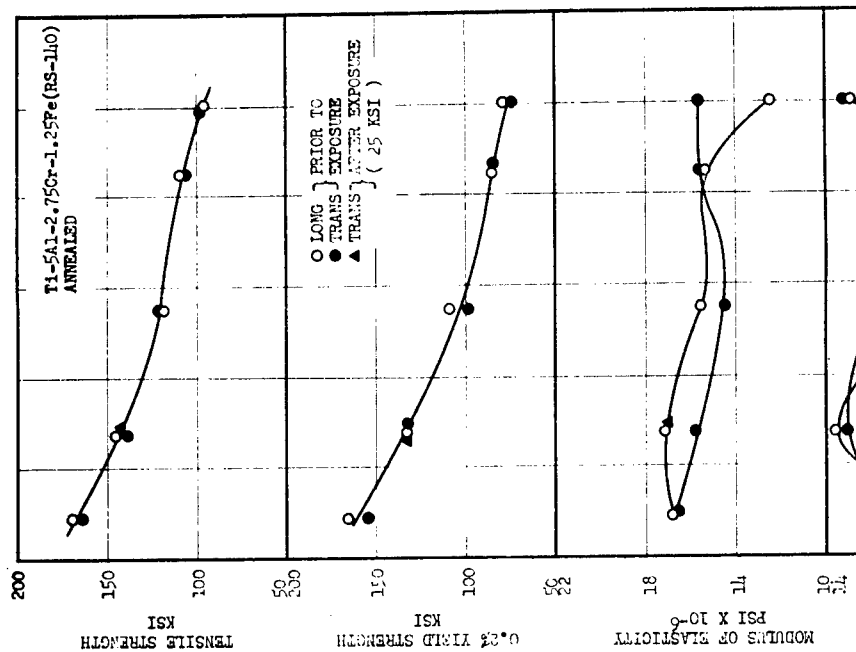


FIG. 6 THE EFFECT OF TENSILE STRENGTH, THE MODULUS OF ELASTICITY AND THE ELONGATION IN PERCENT OF AL-1 - 201 SOLD REDUCED 60% MATERIAL TESTED PRIOR TO EXPOSURE AT 25 KSI IN THE EXPOSED CONDITION.

43



ERRATA SHEET
FINAL REPORT - JUNE 62
NASA
CONTRACT NO. NAS-443

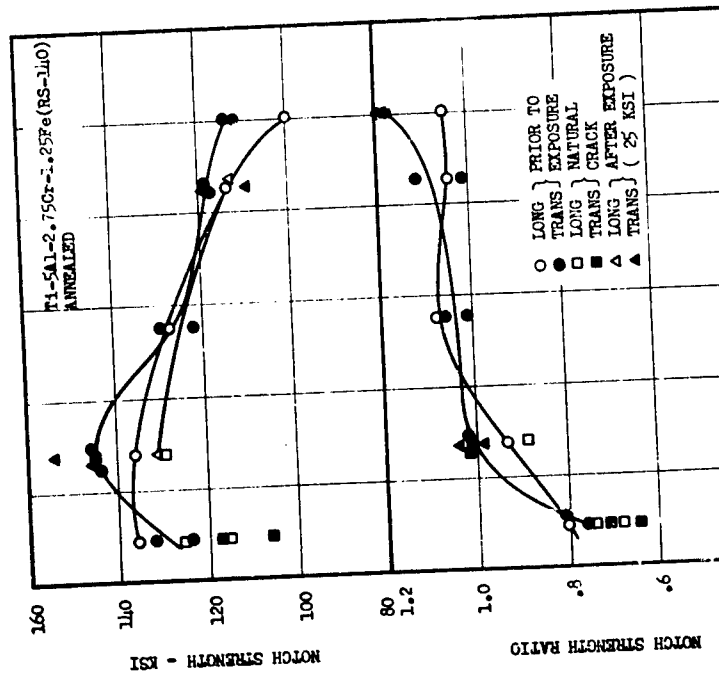
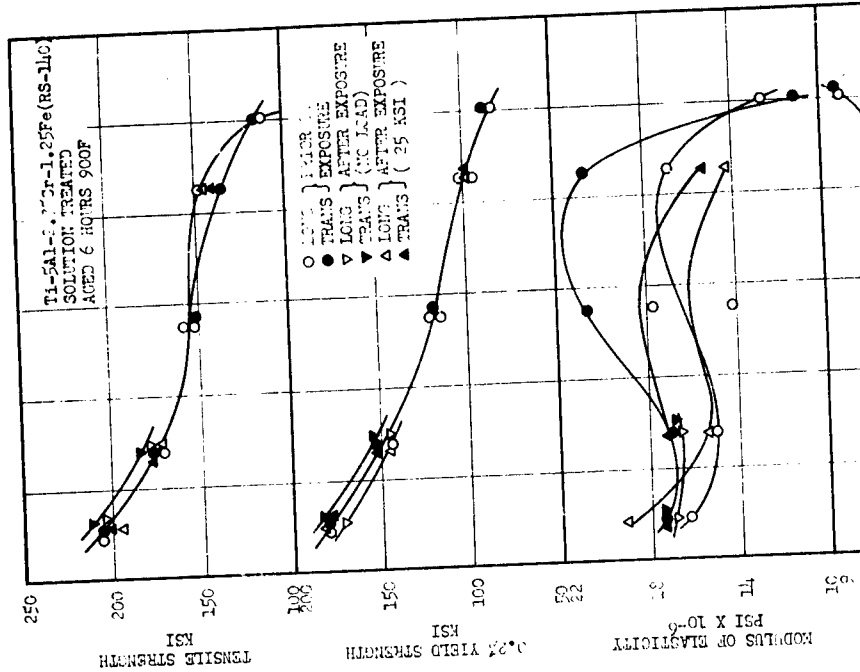
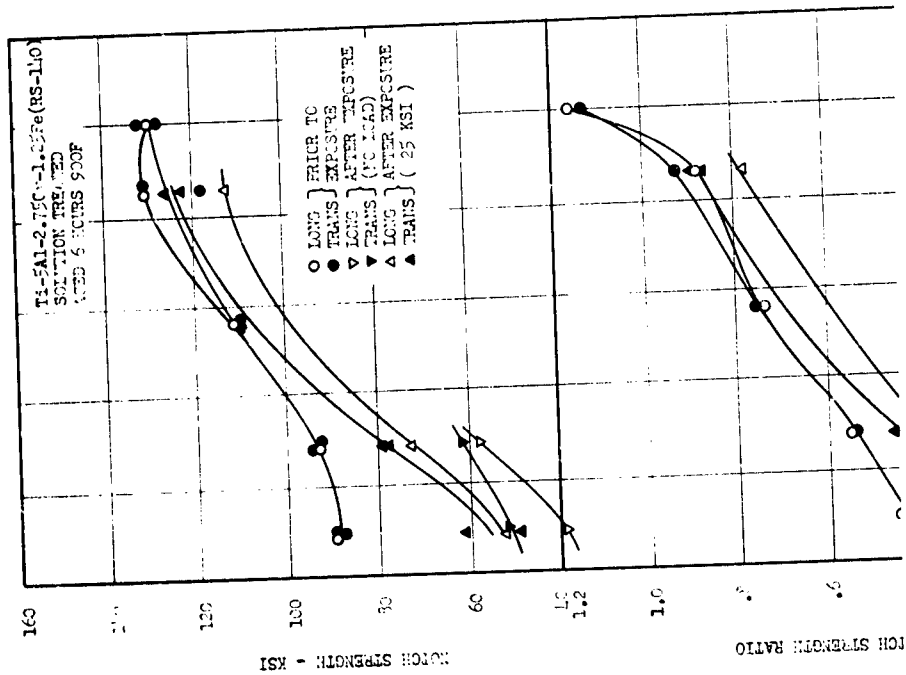


FIG. 8
THE YIELD STRENGTH, TENSILE STRENGTH, AND MODULUS OF ELASTICITY AT THE
ELONGATION IN PERCENT OF 15-7% AND 10-10% CONTRACT NO. NAS-443
MATERIAL TESTED AFTER 10 HOURS OF EXPOSURE AND IN THE EXPOSED
CONDITION.



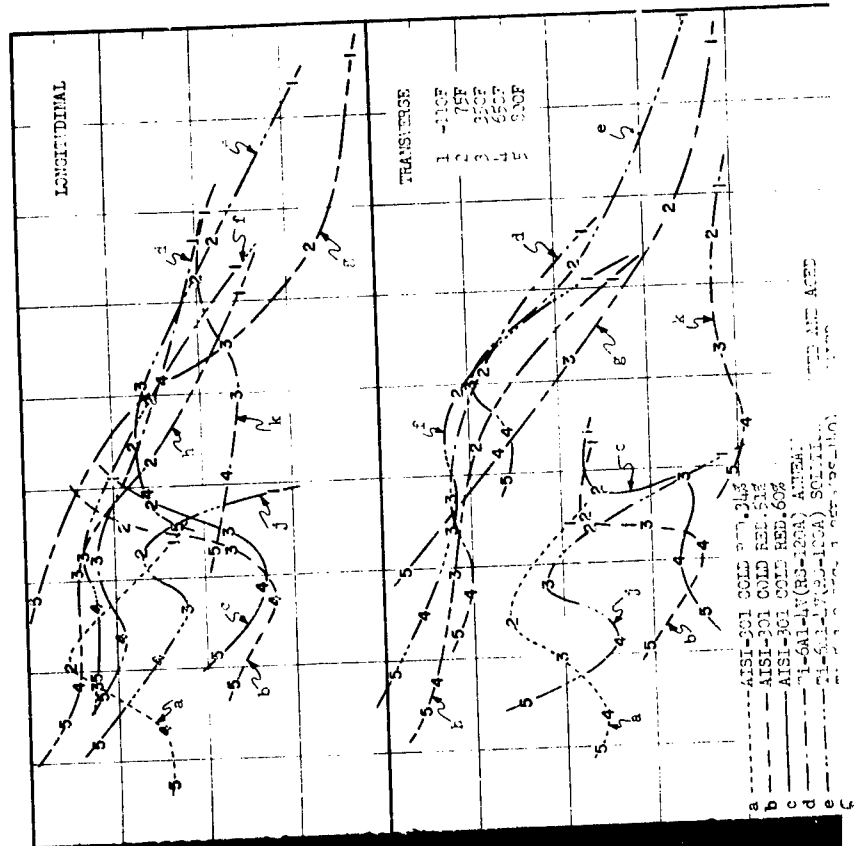
RE 1050. MATERIAL TESTED PRIOR TO EXPOSURE AND IN THE EXPOSED CONDITION.

146



ELONGATION IN PERCENT OF RE 1100 IN CONDITION CH-900.
MATERIAL TESTED PRIOR TO EXPOSURE.

147



THE YIELD STRENGTH, THE MODULUS OF ELASTICITY AND THE
 ELONGATION IN PERCENT OF TI-6AL-4V (RS-120-4) ANNEALED.
 MATERIAL TESTED PRIOR TO EXPOSURE AND IN THE EXPOSED CONDITION.

49

END

48

